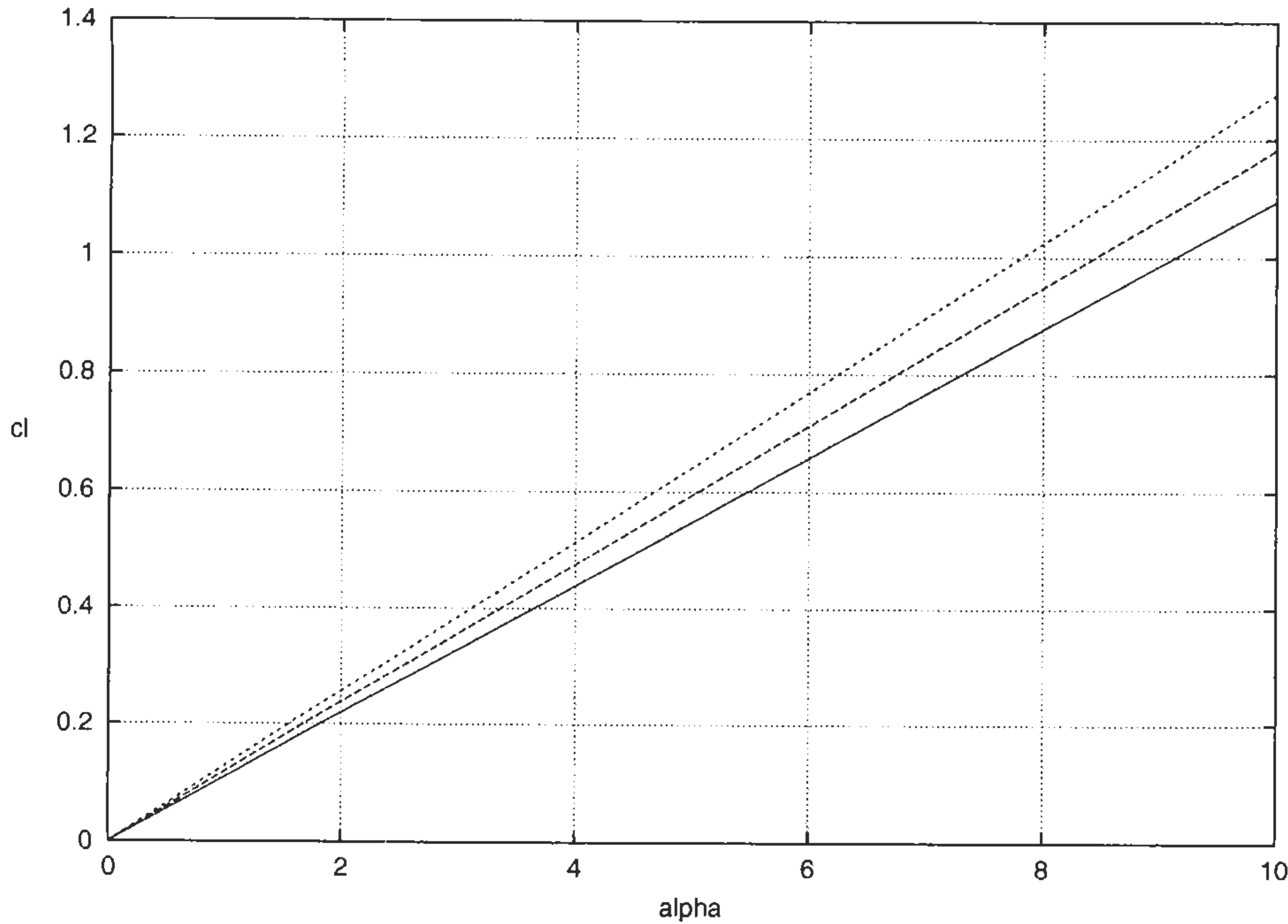


F2a.

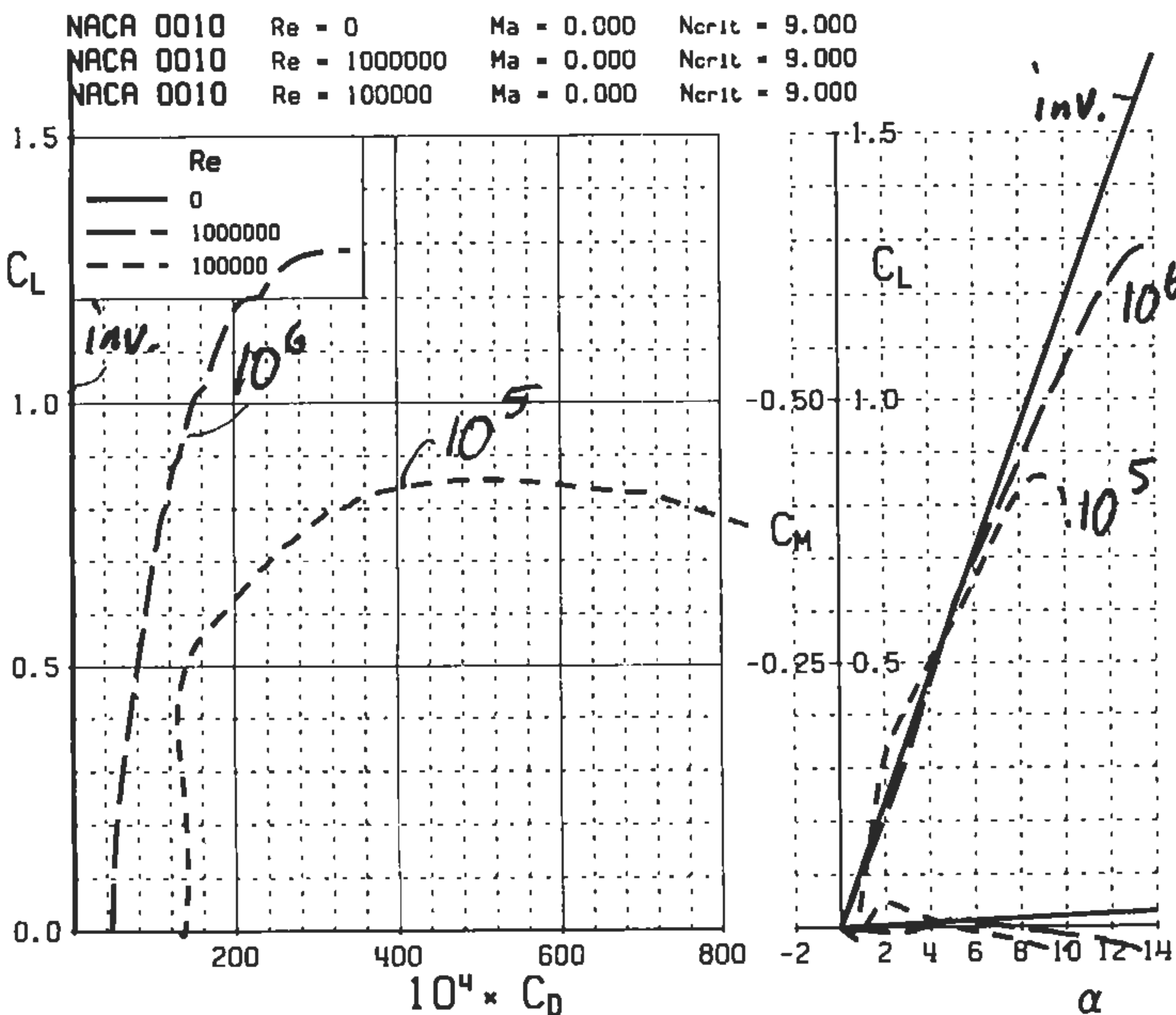


$$C_L = 2\pi\alpha$$

$$= 2\pi\alpha^\circ \cdot \frac{\pi}{180}$$

As airfoil thickness goes to zero, the panel results approach the result from thin airfoil theory (assumptions become more valid).

F2b.



Viscous results approach inviscid results as Re increases (as viscosity gets smaller)